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WIND-TUNNEL RESEARCH COMPARING LATERAL CONTROL DEVICES, PARTICULARLY AT HIGH ANGLES OF ATTACK

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IX. TAPERED WINGS WITH ORDINARY AILERONS

By Fred E. Weick and Carl J. Wenzinger Langley Memorial Aeronautical Laboratory

> Washington February, 1933

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WIND-TUNNEL RESEARCH COMPARING LATERAL CONTROL DEVICES, PARTICULARLY AT HIGH ANGLES OF ATTACK IX. TAPERED WINGS WITH ORDINARY AILERONS By Fred E. Weick and Carl J. Wenzinger

SUMMARY

This report is the ninth on a series of systematic tests in which various lateral control devices are compared, with particular reference to their effectiveness at high angles of attack. The present tests were made with ordinary flap-type ailerons on two wings with different amounts of taper, one medium and the other extreme. On each wing both medium-sized tapered ailerons and short wide tapered ailerons were tested and, in addition, on the wing with the extreme taper, medium and short wide ailerons having a constant chord were tested.

The tests, which were made in the N.A.C.A. 7 by 10 foot wind tunnel, showed the effect of the different plan forms on the general performance and lateral stability characteristics of the wings, as well as the effect of the different aileron shapes on the lateral controllability. It was found that the rolling control given by the ailerons on the wing with medium taper was about the same below the stall as that for corresponding ailerons on rectangular wings, but above the stall the rolling control was somewhat lower than on rectangular wings and well below an assumed satisfactory value. At angles of attack below the stall the yawing moments caused by the ailerons were somewhat lower on the wing with medium taper than on a rectangular wing, but just above the stall the adverse yawing moments were greater. The ailerons on the wing with extreme taper gave better lateral control at angles of attack below the stall in regard to rolling, yawing, and hinge moments than corresponding ailerons on rectangular wings or on the wing with medium taper, but just above the stall the rolling moments fell off almost completely and adverse yawing moments of great magnitude occurred.

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INTRODUCTION

A series of systematic wind-tunnel investigations, one of which is covered by this report, is being made by the National Advisory Committee for Aeronautics in order to compare various lateral control devices. The various devices are given the same routine tests to show their relative merits in regard to lateral controllability and their effect on the <u>lateral</u> stability and on airplane performance. They are being tested first on rectangular Clark Y wings of aspect ratio 6, and then on wings with different plan forms and also wings with such variations as washout and sweepback, which affect lateral stability.

Fart I of this series (reference 1) dealt with three different sizes of ordinary ailerons on rectangular wings. One of these allerons was of medium size taken from the average of a number of conventional airplanes, one was extremely short and wide, and the other was extremely long and narrow. All the ailerons were proportioned to give approximately equal controllability at angles of attack below the stall with equal up-and-down deflection. The results were analyzed to show the relative merits of the three sizes of ailerons when set in the above manner and also when set with two differential movements, and with upward movement only. The narrow-chord allerons were found to be definitely inferior to the medium and wide ones in regard to rolling moments at the high angles of attack, -

Parts II and III (reference 1) deal with other forms of ailerons and lateral control devices on rectangular wings. Part VIII covers tests of medium and wide conventional ailerons on wings with rounded tips, and the present report deals with conventional ailerons on tapered wings. Model wings with medium and extreme taper were used, the first having the center-chord length five-thirds that of the tip chord, and the second having the centerchord length five times that of the tip chord. Since narrow-chord ailerons had given very low rolling moments at high angles of attack on a rectangular wing, the tapered wings were tested with medium chord and wide chord ailerons only.

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APPARATUS

<u>Wind tunnel</u>.- The N.A.C.A. 7 by 10 foot wind tunnel, which is being used throughout the entire investigation, has an open jet and a single closed return passage. The tunnel, together with the regular balance and associated apparatus, is described in detail in reference 2.

Models .- The tests were made with flap-type allerons on two wings, one wing having a 5:3 taper and the other a 5:1. Both wing models were constructed of laminated mahogany, with spans of 60 inches, aspect ratios of 6, and Clark Y airfoil sections along the entire span. The wings had equal taper of the leading and trailing edges, and the maximum ordinates of all sections were in a horizontal plane on the upper surface. On each wing both mediumsized tapered ailerons and short wide tapered ailerons were tested and, in addition, on the wing with 5:1 taper, medium and short wide ailerons having a constant chord were tested. Inasmuch as previous tests (reference 1) had shown that the moments caused by both right and left ailerons could be found separately and added together to give the total effect of both with a satisfactory accuracy, the present tests were made with the right aileron only. Each wing model was equipped with a removable tip portion as shown in Figures 1 and 2, and a different model of this portion of the wing was made for each of the ailerons.

The tapered ailerons were tapered with the wings, the chord of the medium-sized ones (A, figs. 1 and 2) at any longitudinal section being 25 per cent of the wing chord at the same section, and the chord of the short wide ones (B, figs. 1 and 2) being 40 per cent of the wing chord at any section. The ailerons with constant chord (C and D, fig. 2) had the same chord dimension, as the average chord of the tapered ailerons on the wing with 5:1 taper. These constant chord ailerons on the tapered wing were of the nature of skewed ailerons on rectangular wings. The aileron spans were all selected to give approximately the same rolling control at angles of attack below the stall as the medium ailerons on a rectangular wing. (Part I, reference 1.)

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Section 1

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TESTS

The tests were conducted in accordance with the standard procedure, and at the dynamic pressure and Reynolds Number employed throughout the entire series of investigations on lateral control. (Reference 1.) The dynamic pressure was 16.37 pounds per square foot, corresponding to an air speed of 80 miles per hour at standard density, and the Reynolds Number was 609,000, based on the average chord.

TABLE I

SIMULTANEOUS AILERON DEFLECTIONS WITH ASSUMED DIFFERENTIAL MOVEMENTS -----

المريطي المراجع التي ويتراجع يتراجع الأي وير Angles Measured about Aileron Axis

Average diffe	rential (No. 1)	Extreme diffe:	rential (No. 2)
Upward displacement	Downward displacement	Upward displacement	Downward displacement
Degrees	Degrees	Degrees	Degrees
0	0	Ö	0
10	8.5	10	7
20	- 13	20	12
30	15	30	14
35	15	40	11,5
		50	. 7

The regular force tests were made, at 0° yaw, with a sufficient number of angles of attack to determine the maximum lift coefficient, the minimum drag coefficient, and the drag coefficient at $O_{\rm L} = 0.70$, which is used to give a rate-of-climb criterion. Because of the large effect of yew on the lateral stability, tests were made not only at O yaw, but also with an angle of yaw of 20°, which represents the conditions in a fairly severe sideslip. Free-autorotation tests were made to determine the angle of attack above which autorotation was self-starting with ailerons neutral. Forced-rotation tests were also made in which the rolling moment while rolling was measured at the rotational velocity corresponding to $\frac{p^{\dagger}b}{2} = 0.05$, the highest rate likely to be obtained in gusty air, and at angles of yaw of both 0° and $+20^{\circ}$.

<u>Aileron movements</u>.- From tests with the single ailerons deflected upward and downward various amounts, data were obtained from which the results were computed for four aileron movements: the equal up-and-down, average differential, extreme differential, and up-only movements. These movements were the same as those used in Part I. (Reference 1.) The relative up-and-down displacements with the two differential movements are given in Table I and the assumed linkages to obtain all of the movements in Figure 3. The deflection of the ailerons was measured in a plane perpendicular to the hinge axis, and is slightly greater than the projected angle of deflection in a longitudinal plane.

<u>Accuracy</u>.- The accuracy of the results presented in this report is the same as that obtained in Part I. It is considered satisfactory at all angles of attack except in the burbled region between 20° and 25° when the rolling and yawing moments are relatively unreliable due to the critical, and often unsymmetrical, condition of the burbled air flow around the wing.

RESULTS

<u>Coefficients.</u> The force-test results are given in the form of absolute coefficients of lift and drag and of the rolling and yawing moments:

 $C_{L} = \frac{\text{lift}}{q \text{ S}}$ $C_{D} = \frac{\text{drag}}{q \text{ S}}$ $C_{l'} = \frac{\text{rolling moment}}{q \text{ b S}}$ $C_{n'} = \frac{\text{yawing moment}}{q \text{ b S}}$

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where S is the total wing area, b is the wing span, and q is the dynamic pressure. The coefficients are obtained directly from the balance and refer to the wind (or tunnel) axes. In special cases in the discussion where the moments are used with reference to body axes, the coefficients are not primed. Thus the symbols for the

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rolling and yawing moment coefficients about body ares are C1 and Cn. The results as given are not corrected for tunnel-wall effect. an ann an Anna an Anna

The results of the forced-rotation tests are given, also about the wind axes, by a coefficient representing the rolling moment due to rolling:

where
$$\lambda$$
 is the rolling moment measured while the wing is
rolling, and the other factors have the usual significance.
This coefficient may be used as a measure of the degree of
lateral stability or instability of a wing under various
rolling conditions. In the present case, it is used to
indicate the characteristics of a wing when it is subject-
ed to a rolling velocity equal to the maximum likely to be
encountered in controlled flight in very gusty air. This
rolling velocity may be expressed in terms of the wing
span as

$$\frac{p'b}{2V} = 0.05$$

where V is the air speed at the center section of the wing, and p' is the angular velocity in roll about the wind axis.

Tables .- The results of the tests are given in Tables II to XV. Table II gives values of CL, CD, Cl', and Cn' for all aileron deflections (one aileron only) at $\tilde{0}$ yaw for the wing with 5:3 taper, and medium aileron. Table III contains similar data for the same wing and aileron combi-nation, but with -20° yaw. Tables IV and V are similar to II and III, but contain the data for the short wide ailerons on the same wing. Table VI contains the results of the rotation tests for the same wing. Similarly, Tables VII to XY give the results for the wing with 5:1 taper.

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DISCUSSION IN TERMS OF CRITERIONS

For a comparison of the different lateral control arrangements, the results of the tests are discussed in terms of criterions, which are explained in detail in Part I and briefly in the following paragraphs. By use of these criterions a comparison of the effect of the different control devices on the general performance, the lateral controllability, and the lateral stability may be made. The values of the criterions summarizing the results of the present tests are given in Table XVI, and the values for the standard and the short, wide ailerons of Part I (rectangular wings) are included for comparison.

General Performance

(Ailerons Neutral)

<u>Wing area required for desired landing speed.</u> The value of the maximum lift coefficient is used as a criterion of the wing area required for the desired landing speed, or conversely for the landing speed obtained with a given wing area. The value of the maximum lift coefficient was nearly the same for the tapered wings as for the rectangular, but the wing with 5:3 taper had a very slightly higher value than the rectangular wing, and the wing with 5:1 taper had a very slightly higher value than that with 5:3 taper.

<u>Speed range.</u> The ratio $C_{\rm Lmax}/C_{\rm Dmin}$ is a convenient figure of merit for comparison of the relative speed range obtained with various wings. The value of the speed-range ratio was slightly greater for the wing with 5:3 taper than for the rectangular wing, and was still greater for the wing with 5:1 taper. It was about the same for the wing with 5:1 taper as for the straight wing with long rounded tips tested in Part VIII. (Reference 1.)

<u>Rate of climb.</u>- In order to establish a suitable criterion for the effect of the wing and the lateral control devices on the rate of climb of an airplane, the performance curves of a number of types and sizes of airplanes were calculated, and the relation of the maximum rate of climb to the lift and drag curves was studied. This in-

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vestigation showed that the L/D at $C_{\rm L} = 0.70$ gave a consistently reliable figure of merit for this purpose. The numerical value of this criterion was slightly lower for the wing with extreme taper than for the wings with either 5:3 tapèr or réctangular form.

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n na service de la companya de la co La companya de la comp Rolling criterion. - The rolling criterion upon which the control effectiveness of each of the aileron arrangements is judged is a figure of merit which is designed to be proportional to the initial acceleration of the wing tip that follows instantaneous deflection of the allerons from neutral, regardless of the air speed or the plan form of the wing. Expressed In coefficient form, this rolling criterion is

 $\overline{RC} = \frac{C_{I}S_{P}^{2}}{12C_{I}I_{x}}$

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where Cy is the coefficient of rolling moment due to the ailerons with respect to the body axes (which axis for the wing alone is taken as the midspan chord line), and I, is the grea moment of inertia of the wing about the midspan chord line. A more detailed explanation of the derivation of RO and the assumptions upon which it is based is given in Part I, reference 1. and a second second

The numerical value of this criterion that is assumed to represent satisfactory control conditions is approximately 0.075, the value given by the standard ordinary ailerons with the assumed maximum deflection of $\pm 25^{\circ}$ at an angle of attack of 10° . (See Part I, reference 1.)

The comparison of the criterions for the various allerons and movements is given in Table XVI for four representative angles of attack: $\overline{0}^{\circ}$, 10° , 20° , and 30° . The 0° angle represents the high-speed attitudes; $\alpha = 10^{\circ}$ represents the highest angle of attack in which entirely satisfactory control with ordinary ailerons is obtained; $\alpha = 20^{\circ}$ is the condition of greatest lateral instability and is probably about the greatest obtainable angle of attack in a

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steady glide with most present-day airplanes; and finally, $\alpha = 30^{\circ}$ is given only for a comparison with controls for possible future types of airplanes.

At $\alpha = 0^{\circ}$ all the ailerons give values of RC greatly in excess of that considered necessary, the values for the wing with the 5:1 taper being about one-third higher than those for the wings with the 5:3 taper or rectangular forms.

At $\alpha = 10^{\circ}$ the ailerons on the wing with 5:3 taper, as well as those on the rectangular wings, gave values of RC reasonably close to the assumed satisfactory value, but the ailerons on the wing with the 5:1 taper all gave values substantially higher - on the average about one-third higher. Thus, all the ailerons on the wing with 5:1 taper had spans too great, although they were proportioned, to give the same rolling control as the medium ailerons on the rectangular wing at angles of attack below the stall. This condition favors the ailerons on the wing with extreme taper in the comparisons of Table XVI, but inasmuch as these ailerons, even with their large size, give very poor control moments at high angles of attack, the comparison serves the purpose of the present investigation reasonably well.

At $\alpha = 20^{\circ}$ the ailerons on the wing with 5:3 taper gave definitely lower values of RC than the corresponding ailerons on rectangular wings, and the values for the aile-, rons on the wing with 5:1 taper were in most cases so low as to make these ailerons useless for lateral control. The short wide ailerons with both the extreme differential and the up-only movements gave the highest values, those for the tapered ailerons with constant percentage chord being higher than those for the straight ailerons having con-stant absolute chord, but the highest was only about 60 per cent of the assumed satisfactory value. None of the allerons gave moments of useful magnitude with the more conventional equal up-and-down and ordinary differential movements. These tests indicate that ailerons on tapered wings give excellent rolling-control moments at angles of attack below the stall, but that these moments decrease very ranidly as the stalling angle is exceeded so that the control 1990-1997 - Film above the stall is very poor. - - --

At $\alpha = 30^{\circ}$ the ailerons on the tapered wings gave higher values of RC than those on the rectangular wings, but for the wing with 5:1 taper this fact means little, for the values were very low and in some cases negative for the -----

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angles of attack between that for the stall and 30°.

Lateral control with sideslip.- If a wing is yawed appreciably, a rolling moment is set up that tends to raise the forward tip. The magnitude of this rolling moment is always greater at very high angles of attack than the available rolling moment due to ordinary ailerons. The highest angle of attack at which the aileron can balance the rolling moment due to 20° yaw is tabulated for all the arrangements tested as a criterion of control with sideslip. As previously mentioned, 20° yaw represents the conditions in a fairly-severe sideslip. The rolling control against the effect of 20° sideslip for any of the ailerons on the tapered wings was from 1° to 3° lower than for the corressponding ailerons on rectangular wings.

Yawing moment due to ailerons .- The desirable yawing - moment due to allerons depends to some extent upon the type of airplane that is being considered. It is obvious that a yawing moment tending to retard the high wing when the air-plane is banked is never desirable. For highly maneuvorable .military or acrobatic machines, complete independence of the controls as they effect turning moments about the various body axes is probably a desirable feature. On the other hand, at high angles of attack a yawing moment of the proper magnitude tending to retard the low wing would, under certain circumstances, be an appreciable aid to safe flying for large transport airplanes or for machines to be operated by relatively inexperienced pilots. The yawing moments caused by the ailerons on the wing with 5:3 taper vere slightly smaller below the stall than those for the corresponding ailerons on rectangular wings, but just above the stall at an angle of attack of 20° the adverse yawing moments were greater than for the corresponding ailerons on rectangular wings. In fact, for all the aileron deflections except the up-only, the adverse yawing moments above the stall were greater than could be overcome by an average rudder.

On the wing with 5:1 taper at an angle of attack of 0° the ailerons produced smaller values of the yawing moment coefficient than the ailerons on either the rectangular or 5:3 tarered wings, and they produced no adverse yawing moments of serious magnitude. At $\alpha = 10^{\circ}$ no adverse yawing moments of appreciable magnitude were produced by any of the ailerons on the wing with 5:1 taper, regardless of the form of movement. Just above the stall, at $\alpha = 20^{\circ}$, however, all the ailerons with all of the movements, the values

being from three to four times those produced by an average rudder.

Lateral Stability

(Ailerons Neutral)

<u>Angle of attack above which autorotation is self-</u> <u>starting.</u> This criterion is a measure of the range of angles of attack above which autorotation will start from an initial condition of practically zero rate of rotation. The limiting angle of attack was 3⁰ lower for both of the tapered wings than for the rectangular wings.

Stability against rolling caused by gusts. Test flights have shown that in severe gusts a rolling velocity such that $\frac{p!b}{2} = 0.05$ may be obtained. Consequently, the rolling moment of a wing due to rolling at this value of $\frac{p!b}{2}$ gives a measure of its stability characteristics in rough air. In the present case, the angle at which this rolling moment becomes zero is used as a more severe driterion than the previously mentioned angle at which autorotation is self-starting, to indicate the practical upper limit of the useful angle-of-attack range. With 0° yaw the angle of attack for initial instability is also 3° lower for either of the tapered wings than for the rectangular. The actual value of the limiting angle is 14°, which it is interesting to note is 2° below the angle of attack for maximum lift. With 20° yaw the limiting angle for the wing with 5:3 taper was about the same as that for the rectangular wings, but for the wing with 5:1 taper the limiting angle was 3° higher, and had the same value as for 0° yaw.

The above criterion shows the critical range below which stability is such that any rolling is darped out, and above which instability exists. The criterion, maximum C_{λ} , indicates the degree of this instability. With O^o yaw both of the tapered wings had maximum values of C_{λ} which come within the range found for various rectangular wings tested. The range of these values is fairly wide because they depend in a very critical manner on the exact dimensions of the airfoil, and are affected by variations which are well within the ordinary limits of accuracy of construction for wing models.

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The maximum autorotational moment with 20° yaw is of importance only in the condition in which the airplane is skidded and the forward wing tip is rolled upward or the rear tip downward by a gust. With 20° yaw the value for the wing with 5:3 taper was about the same as those for the rectangular wings, but with the wing having the 5:1 taper this autorotational moment had only one-half the value of those for the other wings.

those for the other wings. Control Force Required

The hinge moments of the ailerons on the tapered wings were not measured in this investigation but were computed from the results of previous tests on hinge moments. Using data from reference 3 as a basis, the effect of wing taper on the hinge moments of the required shapes of ailerons was determined, assuming that the hinge moments varied as the square of the aileron chord and directly as the aileron span. The hinge moments of the ailerons on rectangular wings, reported in Part I, reference 1, were computed from reference 4, since those tests were made on similar wings under similar test conditions. The actual hinge moments of the ailerons on the present tapered wings were calculated using the moments of the ailerons on the rectangular wings as a standard, and the effects of taper as determined by the above method.

A coefficient representing the force required on the control stick was then computed in accordance with the following formula:

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$$\mathbf{CF} = \frac{\mathbf{F} \times \mathbf{l}_{2}}{\mathbf{q} \times \mathbf{c} \times \mathbf{S} \times \mathbf{C}_{L}}$$

where **F** is the control force required, and **k** represents the length of the control lever. Similarly to the rolling criterion, the C₁ in the denominator gives the value of the coefficient the proper relation regardless of the angle of attack or air speed, steady flight being assumed. Although the tests described in reference 3 were made at a relatively low Reynolds Number, the control forces as computed are believed to be accurate within about 10 per cent. Values of the coefficient are given in Table XVI at 0° and 10° angle of attack for the assumed maximum alleron deflections, the top of the control stick being given the same maximum travel in all cases.

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It will be noted that the control forces for ailerons on the wings tapered 5:3 are reduced by about 32 per cent of the values for corresponding ailerons on rectangular wings. The control forces for the tapered-chord ailerons on wings tapered 5:1 are reduced by about 57 per cent and the forces for the constant-chord ailerons are reduced by about 65 per cent of the values for the corresponding ailerons on rectangular wings.

CONCLUSIONS

1. The general performance of the tapered wings was slightly better than for the rectangular in regard to speed range, but was slightly poorer in regard to climb, the effects being greater for the wing having 5:1 taper than for that with 5:3 taper.

2. The rolling control given by the ailerons on the wing with 5:3 taper was about the same below the stall as that for corresponding ailerons on rectangular wings, but above the stall it was somewhat lower than for the rectangular wings, and also well below the assumed satisfactory value. At the angles of attack below the stall, the yawing moments caused by the ailerons were somewhat lower than with the rectangular wings, but just above the stall the adverse yawing moments were greater.

3. The ailerons on the wing with 5:1 taper gave better lateral control at angles of attack below the stall in regard to rolling, yawing, and hinge moments than the corresponding ailerons on rectangular wings or on the wing with 5:3 taper, but just above the stall the rolling moments fell off almost completely and adverse yawing moments of great magnitude occurred.

4. The autorotational tendencies of both tapered wings were about the same magnitude as those of the rectangular wings, but started at an angle of attack about 3° lower than for the rectangular wings and about 2° below that for maximum lift coefficient.

Langley Memorial Aeronautical Laboratory, National Advisory Committee for Aeronautics, Langley Field, Va., November 16, 1932.

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n	40°			+	.006	<u> </u>	00	; 	.06	: 	.057		.048	008	.037	.012	.01	30	15		
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J.	000	0.010	0.074	0.145	0.358	0.716	1.055	1.158	1.347	1.273	1,282	1.345	1.180	1.105	1.033	0.740	0.755	0.750	0.693	0.5
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							_			TA	BLE Y									
	- <u></u>					WI	TH IAI	PERED (STS.	TAI OLARK BHORT	T WING	TAPE	CONTE A	3 LILEP.CI	I OMLY)				
-	α Is	_5°	-4°	-3 ⁰	00	₩1 5 ⁰	TH IAI	PERIED (STS. DEORD 1	TAI OLARK BHORT 15°	Y WIRG Y WIRG MIDE AN 16°	TAPE LERON	18°	3 ILLEP.OJ 20 ⁰	83 ₀ 1 Ohra	85°	30°	40°	50 ⁰	60
	α δ _A	_5°	-4°	-3°	0°	¥1 5 ⁰ 0.639	TH 141	ILOAR	14° 1.122	TAU OLARE BHORT 15° M and 1,150	Y WING Y WING NIDE AN 16° neutra	17°	18° 18° 18° 30° ya	300 300 1.170	1.128	85°	30 ⁰	40°	50°	60
	δ	-5° 0.003 .017 .005	-4°	-3° 0.130 .017 .006	0.318 .031 .008	50 0.639 .041 .010	10° 10° 0.940 .077 .015	12° 12° 11eron 1.048 .093 .015	14° 1.122 1.123 .113 19	TA OLARE BHORT 15° 15° d and 1.150 .120 .023	I WING I WING I WING I MING I	17° 17° 1.178 .143 .047	16° 16° 16° 16° 1.178 .160 .058	30° 30° 1.170 .208 .103	r only 22° 1.128 .963 .117	25° 0.853 .403 .094	30 ⁰ 0.827 .485 .075	40 ⁰ 0.785 .658 .053	50°	60 0.6 1.0
	δ_ 0° 0° 0° 0°	_5° 0.003 .017 .005 008	-4°	-3 ⁰ 0.130 .017 .008 002	0.318 .031 .008 003	50 0.639 .041 .010 003	10 ⁰ 10 ⁰ 0.940 .077 .015 005	2000 TI 2001 TI 200	14° 1 looka 1.122 .113 .019 008	TAI OLARIX BEORT 15° d and 1.150 .120 .023 010	160 160 160 1.161 .130 .032 011	17° 17° 11 1.176 .143 .047 018	18° 18° 18° 1.178 .160 .059 014	30° 1.11270 .208 .103 020	1.128 .263 .117 026	25° 0.853 .403 .094 040	30 ⁰ 0.827 .485 .075 042	40° 0.785 .658 .053 040	50° 0.727 .839 .049 045	60 0.60 0
	δ. 0° 0° 0° 0° 0° 0° 0° 0°	-5° 0.003 .017 .005 003	-4°	-3° 0.130 .017 .008 002	0° 0.318 .031 .008 003	50 0.639 .041 .010 003	FC TH 141 10° 0.940 .077 .015 005	13° 13° 1100 1.048 .093 .015 006 111eror 1.025	140 1.122 .113 .019 .008 2.100k 2.100k	TAN OLARE BEORT 15° 16 and 1.150 .023 010 ad and 1.130	ELE V Y WINC WIDE AT 16° neutrs 1.161 .150 .032 011 neutrs 1.185	17° 17° 11 1.176 .143 .047 018 11	16° (0111 20° yr 1.178 .160 .059 014 -20° yr 1.168	3 30° 1.170 .208 .103 020	1.128 .22° 1.128 .263 .117 026	25° 0.853 .403 .094 040	30° .827 .485 .075 ~.042	40° .658 .053 040	50° 0.727 .839 .049 045	60 0.6 1.0 0
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		-5° 0.002 .007 .005 003 .008 .008 .003	_4°	-3° 0.130 .017 .008 002 .017 007 .001	0° 0.318 .021 .008 003 .031 009 .001 0.038 .004 .004	*1 5° .041 .010 003 .041 .041 .041 .003	TH 141 10° 0.940 .075 005 .076 015 .005 0.037 005 .005	ROE TI TRED (12° 11eron 1.048 .092 .015 008 11eron 1.025 .098 019 .006 Ri	14° 14° 1.123 .113 .019 .008 2.109 .008 2.109 .008 2.109 .008 2.109 .008 2.109 .008 .009 .004 .005 .0	TAU OLARE BEORT 15° ed and 1.150 .023 010 ed and 1.130 .118 037 .010 illeron	160 160 160 1.160 .150 .032 011 neutre 1.155 .189 033 .011 Up 0.035 007 .07 .054	17° 17° 11 1.178 .143 .047 018 1.156 .139 034 .0135 -30° 2.036 008	110 5 (OFF 20° yr 1.178 .160 .059 014 .158 .158 .158 .158 .048 .013 /AW 0.037 008	30° 30° 1.170 .108 020 1.170 .208 020 1.170 .208 .015 .015	0.023 0.	25° 0.853 .403 .064 040 0.843 .398 101 0.029 030 .045 .045	30° 0.827 .485 .075 .042 0.850 .485 .075 .043 0.850 .485 .075 .041 0.040 012 .020	40° 0.785 .656 .053 040 0.783 .653 051 .039 0.001 - 006 0.001	50 ⁰ 0.727 .649 045 0.738 .861 048 .045	60 0.60 0 0.6 1.0 0
	0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0°	-5° 0.002 .017 .005 003 .008 .008 .008	-4°	-3° 0.130 .007 .002 .017 .001	0° 0.318 .031 .008 ~003 .031 .003 .001 0.038 .004 .054 .004 .054 .009 .061 .015	• • • • • • • • • • • • • • • • • • •	TH 14 10° 10° 10° 10° 10° 10° 10° 10°	DECE TI DERED (12° 11 eror 1.048 .092 .015 008 11 eror 1.025 .092 008 Ri	ETS. DEORD 1 14° 1.123 .113 .129 .006 2.109 .002 1.099 .024 0.037 .024 0.037 .005 .054 .005 .005 .005	TAI OLARE BHORT 15° ad end 1.150 .120 .023 010 ad end 1.130 .118 037 .010 tleron	1.16° 1.16° 1.16° 1.161 1.130 .032 011 1.155 .189 033 .011 Up 0.038 .001 Up	17° 17° 17° 11 1.176 .1437 013 1.156 .139 034 .015 036 008 008	16° 16° 20° yr 1.178 .160 .014 .188 .014 .188 .014 .188 .015 .048 .015 .048 .058 .058 .058 .058 .058	3 30 ⁰ 1.170 .208 .103 .020 .103 .020 .103 .005 .015 0.041 011 .000 .000 .000 .000 .000	0.028 0.	25° 0.853 .403 .094 040 0.843 .398 010 0.843 .398 010 0.029 020 .045 020 .068 .068	30° 0.827 .485 .075 042 0.830 .485 .075 .041 0.040 ~.012 .020 ~.012 .020 ~.019 .042	40° 0.785 .658 .053 040 0.783 .653 051 .039 0.001 008 .007 011 .030	50° 0.727 .839 .049 045 0.728 .861 045	60 0.6 1.0 0 0.6 1.0 0
	δ	-5° 0.003 .017 .005 008 .018 006 .003	40	-3° 0.130 .017 .006 007 .017 007	0.318 .031 .003 .003 .003 .001 .001 .001 0.038 .001 0.038 .001 .004 .004 .005 .005 .005 .015 .015	50 0.839 .041 .010 003 .041 .001 .002	TH 14 10° 0.940 .077 .015 .005 .078 .078 .078 .078 .005	DECE TI TERED (12° 111eron 1.048 .092 .015 008 111eron 1.025 .098 019 .008 R1	14° 14° 10080 14° 1.122 .112 .112 .112 .019 .008 1.09 .008 109 .008 109 .008 109 .008 .09 .008 .09 .008 .09 .008 .004 .008	TA) OLARI BEORT 15° M and 1.150 .023 010 d and 1.130 .118 020 1.130 .010 tlaron	1.16° 1.16° 1.16° 1.161 1.300 .032 011 1.155 .189 031 1.155 .189 031 UP 0.038 007 .054 005 .003 .0091 .0081 .0081	17° 17° 11 1.176 .143 .047 -012 1.156 .139 -034 .013 -30° 2.038 008	16° 16° 16° 110° 100° 1.178 1.178 1.178 1.188 1.188 1.188 1.188 1.188 1.188 1.188 1.188 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.188 1.003 1.003 1.188 1.003 1.188 1.003 1.0	3 11 LEPO1 20° .208 .102 .103 .103 .103 .203 .025 .015 0.041 .011 .080 .012 .090 .007 .108	1.128 .263 .117 026 0.875 .241 026 .028 0.028 0.028 023 .043 .026 .028	25° 0.853 .094 040 0.843 .398 101 .038 0.029 030 .045 028 .028 .028 .028 .028	30° 0.827 485 075 042 0.830 495 073 .041 0.040 013 .020 013 .021 .023 021 .021 .053	40° 0.785 .656 .053 .053 .053 .051 .039 0.001 .039 0.001 .008 .029 .029	50° 0.727 .839 045 .861 045	0.6 1.0 0 0
	0° 0° 0° 0° 0° 0° 0° 0° 35° 0° 35° 50° 80° 80° 80° 80°	-5° 0.003 .017 .005 008 .008 .008	_4°	-3° 0.130 .007 008 009 009 001	0.318 .031 .031 .033 .031 .031 .031 .031 .031	0.639 .041 .010 003 .041 .041 .041 .003	TH 141 10° 0.940 .077 .015 .078 .005 .007 .005 .005 .007 .005 .007 .005 .005 .007	DECE II TERED (12° 11° 1048 - 092 015 - 008 1025 - 009 1025 - 009 1025 - 009 84 84 84	140 1 look 1.122 1.123 1.123 1.123 1.024 1.025 1	TAU OLAFR BHORT 1E ⁶ ad and 1.150 .023 010 d and 1.130 .023 037 .010 lleron	LE Y Y WING FIDE AI 16° neutrs 1.161 .130 011 neutrs 1.185 .189 032 .032 .032 .032 .032 .032 .032 .032 .032 .032 .033 .035 .036 .007 .054 .005 .005 .005 .005 .005 .005 .005 .005 .005 .005 .005	+ TAPEE 17° 11 1.17847012 11563303	18° 18° 20° 1.168 .059 014 .158 .048 .058 .058 .059 .014 .058 <	3 30° 1.170 .208 .103 -020 .303 -020 .303 -041 -011 .065 .015	0.0375 .241 .028 0.029 .024 0.029 .024 0.029 .024 0.026 .021 .024 0.026 .026 .026 .026 .026 .026 .026 .02	25° 0.853 .403 .094 040 0.843 .398 .398 .398 .398 .398 .398 .398 .398 .398 .398 .398 .398 .398 .398 .023 .023 .023 .025 .023 .025	30° .485 .075 .042 0.830 .485 .075 .041 0.042 .020 .049 .049 .049 .049 .049 .049 .049 .04	40° 0.788 .659 .053 040 0.783 .653 051 .039 0.001 051 .009 011 .020 016 .029 016	50 ⁰ 0.727 .839 045 0.728 .861 045	60 0.60 0 0.60 0
	α δ _A 0° 0° 0° 0° 0° 0° 0° 0° 0° 0°	-5° 0.003 .017 -005 -005 -006 -006 -006	_4°	-3° 0.130 .002 002 .017 .001	0° 0.318 .031 .031 .003 .003 .004 .004 .005 .005 .005 .005 .005 .005	50 0.639 .041 .010 .003 .041 .003	TE IAI 10° .0.940 .0.977 .015 .078 .005 .005 .005 .005 .005 .005 .005 .005 .005 .005 .005 .005 .007 .013 .007 .013 .007 .013 .007 .015 .007 .015 .005	PROF T1 PERED (12° 11eror 1.045 .092 .015 .092 .092 .092 .092 .092 .092 .092 .092	STS. JROBD 4 14° 1 look 1.123 .113 .113 .113 .113 .113 .123 .123 .006 .008	TAU OLAFIT 180077 180077 180077 10 30 and 1.150 .023 010 1.150 .023 010 1.150 .023 010 1.150 .023 010 1.150 .023 .010 1.150 .023 .023 .010 1.150 .023 .023 .023 .023 .023 .023 .023 .02	LE Y T TIRC TIDE A1 16° neutrs 1.161 .130 .032 .032 .031 neutrs 1.165 .1393 .011 1.155 .1393 .011 up 0.035 .035	+ TAPEE LLEROY 17° 1 1.178 .143 .047 034 .139 034 .139 034 .139 034 .139 .036 036 036 008 	The second secon	3 30° 30° 30° 30° 300 300 300 300 300 30	(OJLY 220 1.128 .833 .341 036 0.875 .241 036 0.28 0.028	25° .403 .403 .094 040 0.843 .398 -101 .038 0.029 030 .045 035 .080 030 030 030	30° 0.827 .042 .042 .042 .042 .042 .042 .041 .053 .041 .020 .042 .020 .012 .020 .012 .021 .023 .021 .053 .051 .051 .051 .051 .051 .051 .055 .055	40° 0.785 .658 .053 .053 .055 .055 .051 .039	50° 0.727 .839 .049 045 .045	60 1.0.70 0 0.80 0
	δ₄ 0°0°0°0 0	-5° 0.002 .017 -005 008 .018 008	_4°	-3° 0.130 .017 007 .001	0° 0.318 .031 .031 .031 .031 .031 .035 .031 .003 .004 .005 .005 .005 .005 .005	50 0.839 .041 .010 .041 .003 .041 .003	FC 141 10° 10° 10° 0.940 0.940 0.077 .015 .005	DECT II TERED (12° 110 - 12° 10 - 12° 10 - 12° 10 - 12° - 002 - 000 - 002 - 0	STS. JROND 1 14° 1 lookd 1.122 .019 .029	TAU OLAFT 180077 18077 1150 3d end 1.150 .023 010 1.130 037 .010 1.120 1.1	512 Y T WING FIDE AI 16° 1.161 1.161 1.163 1.163 1.163 1.163 1.163 1.163 1.163 1.163 1.163 1.003 0.033 0.033 0.011 0.054 0.005 0.005 1.004 0.005 0	+ TAPE LLEROF 17° 1 1.176 013 034 034 035 0366 036 036 036 036 036 036 036 0	The second secon	3 30° 30° 30° 30° 30° 30° 30° 30° 30° 30	(OJLY 22° 1.128 .663 .617 .241 038 0.039 031 .043	25° 25° 0.853 .094 .023 .094 .085 .094 .085 .094 .086 .028 .038 .028 .038 .028 .038 .028 .038 .028 .038 .048 .038 .038 .038 .038 .038 .048 .038 .048 .038 .048 .038 .048 .038 .048 .038 .0488 .0488 .0488 .0488 .0488 .04888 .0488 .0488 .0488 .048888	30° 0.827 .485 .075 .075 .075 .075 .075 .042 .485 .041 0.640 .032 .032 .041 0.040 .053 .053 .053 .053 .053 .053 .053 .05	40° 0.785 .658 .653 .053 .053 .055 .055 .039 0.001 .039 0.001 .039 .001 .039 .018 .039 .018 .039 .018 .039 .018 .039 .040	50° 0.727 .839 .049 .049 .045	60 0.60 1.0? 0 0.60 1.0?

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						. 1	OTATI	N TEST	5. OL	RE T	TING TI	PERED	5:	3						
					a. 11	. of ver	for	forásá	rotatio		<u>p'b</u>	0.05	٠. {+}	aidir	ug rot	ation				
					×λ -	6210					3 7		()	dampi	ing ro	tation				
	· · · ·				<u> </u>		R.R	. = 809	,000	Velo		- 80 =	.p.n.							
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ol wi	tion cok-	ο _λ	-0.0	23	0:	.90	.80	.00	0.030	.036	.03	5	.000		003	00 5	003	003	002	008
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						. 8791	FORO	TESTS	CLAI		UNG TAN	ERED .	5 : 1 LEROI	K 0873	r)	•				.'
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Ł	°°	-0	.010	.061	0.132	0.336	0.683	1.005	1.114		.278	15781	1.273	1.212	50.97	0 0.77	9 0.71	3 0.718	0.866	0.553
ł	0° 0°		.010 .017	0.061	0.132 .016	0.336 .021	0.683	1.005	1.114 .102 Right	.123 : sile:	.144	1tral 1.288 .162	1.273	1.212	5 0.97 0 .33	00.77	9 0.71 1 .45	3 0.718 8 .651	0.868	0.558
1	0° 0°	0 	.010 .017	0.061	0,132 .016	0.238 .021 .017	0.683	1.005 .081	1.114 .103 Right	.214 .123	.278] .144	1tral 1.288 163	1.273	.004	5 0.97 0 .33	0.77	9 0.71 1 .45	3 0.718 8 .651 4 .004	0.868	0.555
10 v av a	0° 0° 10° 20° 20°	0	.010	0.061	0.132	0.336 .031 .031 .017 .001 .031 .001	0.683	1.005 .081 .081 002 .031 003	1.114 .103 Right			1tral	1.273		5 0.97 .33	0.77	9 0.71 1 .45 00 00	3 0.718 8 .651 4 .004 4004 9 .005 7005	0.866	0.553
- R - R - R - R - R - R - R - R - R - R	0° 0° 10° 10° 20° 20° 25° 25° 25°	0 	.010	0.061	0,132	0.336 .031 .017 001 .031 .036 .003 .003	0.683	1.005 .081 .031 003 .038 003 .058 003	1.114 .103 Right	.038 .004	.037	14ral	.018	.004 003 005 007 007 007	5 0.97 33 4 5 7 01	100	9 0.71 1 .45 00 00 00 0300	3 0.718 8 .651 4 .004 4 .006 7006 3 .013 8011 7 .017	0.866	0.553
	0° 0° 10° 10° 20° 20° 20° 20° 20° 20° 20° 20° 20° 2	0	.010	0.061	0.132	0.236 .021 .017 001 .031 .036 .002 .041 .008 .045	0.683	.014 .031 .031 .031 .035 .040 .040 .040	1.114 .103 Right	.038 .038 .040	.037 .030	1tral 1.288 .162	.018	.004 003 003 007 007 007 007 007 008	5 0.97 .33 5 5 7 01 7 01 5 02	100 500	9 0.71 1 .45 .00 00 .00 00 .01 .01 .01 .01 .01 .02	3 0.718 8 .651 4 .004 4004 7009 5 .013 8011 9013 9013 0 .031	0.866	0.553
	0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0	0	.010	0.061	0.132 .016	0.336 .021 .031 .031 .031 .036 .002 .041 .003 .045 .003 .050 .006	0.683	1.005 .081 .031 003 .035 003 .040 003 .040 003 .045 002 .045 .001	1.114 .103 Right	.038 .038 .004 .003 -	.037 .037 .030 .030	1tral 1.288 1.162	.018 .017 .008 .017 .006	1.212 .220 003 003 005 005 005 005 005 005	5 0.97 .32 .32 .32 .32 .33 .33 01 01 01 01 01 01	100 500 7 .00 500	9 0.71 1 .45 .00 00 0.00 00 0.01 00 1 .02 00 1 .02 00	3 0.718 6 .651 4 .004 9 .005 7 .003 8 .011 7 .013 0 .034 0 .017 6 .017	0.866	0.553
	0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0	Q	.010	0.061	0.132	0.336 .031 .031 .031 .031 .036 .036 .036 .045 .003 .045 .003 .045 .003 .050 .056 .056	0.683	1.005 .081 -003 .035 -003 .045 -003 .045 -003 .045 -003 .050 .050 .050	1.114 .103 Right	.033 .033 .004 .004 .003 .053 .000	.037 005 005 004 004	1tral 1.288 163	.018 .018 .008 .017 .008	1.212 .220 .004 003 .006 007 .014 006 .016 006 .016 .006 006 .006	5 0.97 33 501 701 701 301 301 501 501	100 500 7701 800 7701 800	9 0.71 1 .45 .00 .00 .00 .00 .01 .01 .01 .01	3 0.718 8 .651 4 .004 9 .005 7 .007 8 .011 7 .012 8 .013 0 .031 0 .031 6 .011 7 .012 8 .013 9 .013	0.866	0.555
	0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0	-0	.010 .017	0.061	0.138	0.236 .021 .031 .031 .031 .038 .045 .002 .045 .003 .045 .003 .056 .006 .068 .068 .068	0.683	1.005 .081 -003 -003 -003 -003 -003 -003 -003 -00	1.114 : .103 Right	.038 .038 .038 .004 .004 .003 .005 .000 .000	.0378] .144 ron up .037 .0037 .003 .030 .045 .003 .053 .001	1tral 1.288 .162	.018 .018 .008 .007 .006 .028 .035 .035 .005	1.212 .220 001 001 001 001 001 001 001 001 001 001 001 001	5 0.97 3 33 4 3 701 4 3 501 501 501 501 501	100 500 701 701 701 701 701 701	9 0.71 45 00	3 0.718 8 .651 4 .004 4 .005 7 .005 8 .012 8 .012 9 .012 0 .031 7 .013 6 .011 7 .013 6 .004	0.866 .843	0.555
	0° 0° 10000 00 000 000 000 000 000 000 0		.010	0.061	0.133	0.236 .021 .031 .031 .036 .045 .045 .045 .045 .045 .050 .056 .058 .058 .058	0.683	1.005 .081 003 005 003 005 -	1.114 : .103 Right	.033 .033 .004 - .040 .063 .000 .003 ailerc	.037 .044 .037 .005 .030 .030 .030 .030 .045 .053 .053 .001		.018 .018 .008 .007 .006 .035 .035 .035	1.212 .230 003 005 001 006 .010 006 .010 006 .010 006 .010 006 .010 006 .010 006	5 0.97 3 33 501	100 500 7701 800 7701 800 1300 1300 1300 1300 1401 1500 1500 1500 1600 1701 1700 1701 1800 1900 1900 1900 1000	9 0.71 45 0000 0000 000 000 000 000 000 000 	3 0.718 8 .651 4 .004 4 .005 7 .005 8 .011 7 .012 9 .013 0 .031 0 .031 0 .017 6 .017 6 .013 6 .013 6 .013 6 .013 6 .013 7 .013 8 .004	0.866 .843	0.555
	0° 0 1000 0		.010	0.061	0.132	0.536 .031 001 .031 .031 .036 .041 .045 .045 .045 .045 .056 .056 .056 .056 .056 .056 .056 .05	0.683	1.005 .031 -003 .031 -003 .035 -003 .040 -003 .040 .050 .050 .050 .055 .003 .055 .003 .055 .003 .055 .003 .005 .003 .003	1.114 1.103 Right	.038 .038 .038 .004 .003 .004 .003 .000 .003 .000 .003 .003	.037 .044 .037 .005 .030 .055 .004 .053 .053 .053 .053 .053 .053		.018 .018 .008 .007 .005 .035 .005 .035 .005 .003	1.212 .220 001 002 001 002 001 002 001 002 001 -	5 0.97 .32 .32 .32 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .33 .01 .01 .01 .01 .01 .01 .01 .01	100 700 700 700 700 2401 301 2500 701 2401 301 2500 2701 2401 20	0.71 0.000 <td>3 0.718 6 .651 4 .004 4 .004 4 .004 4 .004 5 .012 8 .011 7 .012 0 .014 2 .017 6 .011 7 .012 6 .011 7 .012 6 .011 7 .012 3 .002 3 .002 3 .002 3 .002</td> <td>0.866</td> <td>0.555</td>	3 0.718 6 .651 4 .004 4 .004 4 .004 4 .004 5 .012 8 .011 7 .012 0 .014 2 .017 6 .011 7 .012 6 .011 7 .012 6 .011 7 .012 3 .002 3 .002 3 .002 3 .002	0.866	0.555
	0° c 10°			0.061	0.132	0.536 .031 .031 .001 .036 .036 .045 .045 .045 .045 .050 .008 .045 .050 .008 .058 .058 .011 .011 .011 .001 .011 .001 .017	0.683	1.005 .081 003 .035 003 .045 003 .045 003 .045 .055 .045 .055 .055 .055 .055 .055	1.114 1.103 .103 Right	.033 .123 .123 .033 .004 .004 .003 .000 .000 .000 .00			.018 .018 .008 .017 006 .028 .005 .035 003		5 0.97 3 0.97 3 0.97 3 0.97 4 5 5 0.97 701 5 0.97 701 7	100 501 77 .00 7701 301 7801 77 .00 77 .00		3 0.718 6 .651 4 004 4 004 9 .013 8 .017 9 .013 8 .011 7 .007 9 .013 8 .011 7 .012 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .0033 10 .0033 10 .0033 10 .0033 10 .0033 11 .0033	0.886	0.555
	0° 0° 100° 100° 10° 10° 10° 10° 10° 10°			0.061	0.132	0.336 .031 001 .031 .001 .031 .036 .041 .045 .045 .045 .045 .045 .045 .045 .045	0.683	1.005 .081 003 .031 003 .035 003 .040 .005 .003 .005 .003 .005	1.114 1 .103 Right	.033 .123 .123 .123 .123 .123 .123 .123 .1		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	.018 008 008 008 005 005 005 005 003	1.212 .230 001 -	5 0.97 32 5 0.97 32 5 0.97 02 5 0.97 02 5 0.97 02 5 0.97 02	100 100 100 100 100 100 .00 .00 .00 .00 .00 .00 .00		3 0.718 8 .651 4 .004 4 .004 5 .012 8 .611 7 .002 8 .012 8 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .003 3 .002 3 .002 3 .003 1 .004	0.866	0.555
	0°0 1			0.061	0.132	0.336 .021 .017 -001 .031 .032 .032 .032 .032 .032 .041 .032 .045 .045 .045 .045 .045 .045 .045 .045	0.683	1.005 .081 003 003 003 003 003 003 003 003 003 003 003 003 015 015 015 015	1.114 1 .103 Right	.033 .033 .033 .004 .003 .000 .003 .000 .003 .003 aller .002	.045 .007 .003 .003	1 V	.018 .018 .008 .008 .005 .005 .005 .005 .005	1.212 222 -000 -000 -000 -000 -000 -000 -0	5 0.97 5 0.97	0 0.77 8 3 100 7 .00 7 .00 500 7 .00 8 .00 1000 1		3 0.718 8 .651 4 .004 4 .004 6 .005 7 .002 8 .011 9 .003 9 .012 9 .012 9 .014 10 .017 10 .017 10 .011 10 .011 10 .002 11 .002 12 .002 13 .002 14 .002	0.866	0.555
	0° 0° 10° 110° 110° 111° 0° 110° 0° 111° 0° 110° 0° 110° 0° 110° 0° 110° 0° 110° 0° 110° 0° 110° 0° 110° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0			0.061	0.132	0.336 .021 .017 .001 .031 .001 .033 .041 .038 .042 .045 .045 .045 .045 .045 .045 .045 .045	0.683	1.005 .081 -003 -003 -003 -003 -003 -003 -003 -00	1.114 1 .103 . .103 . 	.038 .038 : aller .038 : aller .004 .003 .000 .003 .000 .003 .000 .003 .000 .003	.037 .037 .037 .036 .037 .005 .030 .030 .045 .004 .045 .003 .001 .007 .007	1 V	.018 008 008 005 005 005 005 003	1.812 .282 .000 .000 .000 .000 .000 .000 .00	5 0.97 5 0.97	0 0.77 8 3.55 .100 .500 .7 .00 .7 .00 .700 .700 .7 .00 .7 .00 .00 .00 .00 .00 .00 .00 .00		3 0.718 6 .651 4 004 4 004 5 .013 8 .017 9 .013 9 .013 9 .013 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 0 .031 3 .003 3 .003 3 .003 3 .003 1 .004 3 .003 3 .003 3 .003 3 .003 3 .003 4 .003 3 .003 <	0.866	0.555
	0° 0° 110° 110° 110° 1110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 11110° 110° 11110° 110° 11110° 110° 11110° 110° 11110° 110° 11110° 110° 110° 11110° 110° 110° 110° 11110° 110°° 110°			0.061	0.132	0.336 .021 .017 .031 .031 .031 .031 .032 .041 .045 .045 .045 .045 .045 .055 .055 .055	0.683	1.005 .081 003 003 005 003 005 003 005 003 005 003 005 003 005 005 015	Right				.018 018 008 008 005 005 005 005	1.813 .282 .000 -000 .010 .000 .000 .000 .000 .000	5 0.97 5 0.97	0 0.77 8 35 100 500 7 .00 500 7 .00 7 .00 500 7 .00 7	8 0.414 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	3 0.718 8 .651 4 .004 4 .004 5 .012 8 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .013 9 .003 3 .003 1 .004 1 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002	0.866	0.555
	0°0 1000 2			0.061	0.132	0.336 .021 .031 .031 .033 .032 .032 .032 .032 .032 .032 .041 .032 .045 .032 .045 .045 .045 .045 .045 .045 .045 .045	0.683	1.005 .081 003 003 005 003 003 003 003 003 003 003 003 003 013 013 015 003 015 003 015 003 015 003 015 003 015 003 015 003 015 003 015 003 015 003 015 003 -	1.114 1 .103 Right	.033 .033 .033 .033 .004 .003 .005			-018 -008 -008 -008 -005 -005 -005 -005	1.811 .282 -000 -000 -000 -000 -000 -000 -000 -	5 0.97 1 3 5 0.37 6 0.97 7 -00 7 -00 7 -00 8 -00 3 <	0 0.77 8 35 100 500 701 500 701 300 1500 1600 1700 18 -		3 0.718 8 .651 4 .004 -0012 .0012 5 .0012 7 .0073 8 .0117 7 .0073 8 .0117 7 .0123 0 -0142 6 .0117 7 .0126 8 .0012 8 .0023 3 .0023 3 .0023 1 .0031 1 .0043 4 .0023 3 .0023 3 .0023 3 .0023 3 .0023 4 .0023 3 .0023 4 .0023 3 .0023 4 .0023 3 .0023 4 .0023 3 .0023 4 .0023 5	0.866	0.555
	0° 0° 10° 11° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0° 0°			0.061	0.132	0.335 .021 .017 -001 .031 .033 .036 .036 .036 .036 .041 .036 .041 .036 .045 .045 .045 .045 .045 .045 .045 .045	0.683	1.005 .081 -003 -003 -003 -003 -003 -003 -003 -00	1.114 1 .103 Right	.0150 - 015 .1214 1 .122 .033 .004 .005 .000 - .003 .000 .002 .002 .002 .002		1.288) .162		1.812 .222 .2000 .000 .000 .001 .001 .001		0 0 .77 8 3 .27 100 77 .00 77 .00 78 .00 77 .00 70 .		3 0.718 8 .651 4 .004 4 .004 6 .003 7 .003 8 .011 9 .003 9 .012 9 .013 00 -014 6 .017 7 .012 8 .017 9 .013 9 .012 9 .012 9 .012 9 .012 9 .012 9 .012 9 .012 9 .011 10 .002 1 .002 1 .002 1 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002	0.866	0.555
	0° 0° 110000 11000 11000 11000 11000 11000 11000 11000 11000 11000 110000 11000000			0.061	0.132	0.336 .021 .017 .031 .031 .033 .033 .033 .033 .033 .033	0.683	1.005 .081 -003 -003 -003 -003 -003 -003 -003 -00	1.114 1.109				.018 008 008 008 005 005 005 005 005 003 001 001 001 001 002	1.212 .222 .222 .222 .222 .222 .222 .22	5 0.97 5 0.97	0 0.77 8 35 5 -00 7 .00 7 .00 7 -00 7 -00 7 -00 7 -00 7 -00 10 -00		3 0.718 8 .651 4 .004 4 .004 5 .013 8 .013 8 .013 9 .013 9 .013 9 .013 0 .011 7 .002 9 .013 9 .013 6 .011 7 .002 8 .001 8 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002 3 .002	0.866	0.555
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GERED R.F. = 609, 050 00 50 100 120 140 54 A11 00 -0.098 0.388 0.602 0.987 1.073 00 -0.098 0.388 0.602 0.987 1.073 00 -0.013 .032 .609 .608 007 00 -0.013 .032 .609 .608 007 00 -0.013 .032 .609 .608 .008 00 -0.013 .032 .609 .608 .007 00 -0.02 -0.024 002 002 00 .003 .002 004 003 00 .003 .002 003 .003 00 .001 002 003 .003 00 .001 002 003 .003 00 .003 .005 .003 .003 00 .003 .0035 .0035 .0035</td> <td>0.A. Tsohnicsl Note No. 449
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FORCE TESTS. OLARE I VI
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R.N. = 609,000 Veloc
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</td> <td>0.A. Tsohniogl Hote Ho. 440
TABLE J
FORGE TESTS. CLAFE Y VIDE ALLERON (OR
R.R. = 600,000 Velocity = 80 s.
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0° -0.05 0.258 0.659 0.258 1.059 1.159 1.157 1.157 1.557
0° -0.05 0.258 0.659 0.258 1.059 1.058 1.157 1.157 1.157 1.557
0° -0.05 0.258 0.659 0.258 0.072 1.058 1.157 1.157 1.157 1.557
0° -0.05 0.258 0.059 0.058 0.057 1.058 1.058 1.157 1.157 1.157 1.157
0° -0.05 0.258 0.059 0.058 0.071 1.058 1.168 1.155 1.155 1.157
0° -0.050 0.258 0.058 0.058 0.058 0.058 0.058 1.165 1.158 1.157 1.151
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0° -0.05 0.258 0.058 0.054 0.058 0.050 0.0000 0.0000 0.0000 0.0000 0.0000 0.000 0.</td> <td>0.4. Teohnloal Hote No. 449 FALLE X FORCE TESTS. CLARX I WING TAPERED 5 :1. F.H. = 609,000 Velocity = 80 m.p.h. F.H. = 609,000 F.H. = 100 m.p.h. F.H. = 609,000 F.H. = 1100 m.p.h. F.H. = 60</td> <td>0.4. Technical Hote Ho. 449 FARE T FARE TERS. CLART Y WIRE TAPERED 5 : 1 WITE 7.
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H.A.O.A. Technical Note No. 449

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